

IN THE CLAIMS

Please amend the claims as follows:

Claim 1 (Currently Amended) A turbine moving blade comprising a platform having a gas path surface extending in the combustion gas flow direction, and a blade portion erecting on said platform, said gas path surface of platform being coated with a thermal barrier coating, wherein

said thermal barrier coating substantially covering said gas path surface and being is formed so as to go around from said gas path surface of platform to at least a part of the outer peripheral face of said platform.

Claim 2 (Original) The turbine moving blade according to claim 1, wherein a step portion is formed in at least a part of the peripheral edge portion of said platform, and said thermal barrier coating is formed so that it goes around to said step portion and the end face thereof is in contact with the upper face of said step portion.

Claim 3 (Currently Amended) A turbine moving blade comprising a platform, a blade portion erecting on said platform, and a shroud provided at the tip end of said blade portion, a gas path surface extending in the combustion gas flow direction of said shroud being coated with a thermal barrier coating, wherein

said thermal barrier coating substantially covering said gas path surface and being is formed so as to go around from said gas path surface of shroud to at least a part of the outer peripheral face of said shroud.

Claim 4 (Original) The turbine moving blade according to claim 3, wherein a step portion is formed in at least a part of the peripheral edge portion of said shroud, and said

thermal barrier coating is formed so that it goes around to said step portion and the end face thereof is in contact with the upper face of said step portion.

Claim 5 (Currently Amended) A turbine stationary blade comprising a pair of shrouds each having a gas path surface extending in the combustion gas flow direction, and a blade portion held between said shrouds, at least either one of said shrouds being coated with a thermal barrier coating, wherein

said thermal barrier coating substantially covering said gas path surface and being is formed so as to go around from said gas path surface of shroud to at least a part of the outer peripheral face of said shroud.

Claim 6 (Original) The turbine stationary blade according to claim 5, wherein a step portion is formed in at least a part of the peripheral edge portion of said shroud, and said thermal barrier coating is formed so that it goes around to said step portion and the end face thereof is in contact with the upper face of said step portion.

Claim 7 (Currently Amended) A turbine split ring having a gas path surface extending in the combustion gas flow direction, said gas path surface being coated with a thermal barrier coating, wherein

said thermal barrier coating substantially covering said gas path surface and being is formed so as to go around from said gas path surface to at least a part of the outer peripheral face.

Claim 8 (Original) The turbine split ring according to claim 7, wherein a step portion is formed in at least a part of the peripheral edge portion, and said thermal barrier coating is